

**ILRS SLR MISSION SUPPORT REQUEST FORM** (version: December 2015)

**SUBMISSION STATUS:**

- New Submission (default)
- Incremental Submission (accepted only for a follow-on mission; fill-in new information only)  
(provide the reference mission and the date approved by the ILRS: \_\_\_\_\_)

**SECTION I: MISSION INFORMATION:**

**General Information:**

Satellite Name: Lomonosov

Satellite Host Organization: Scobeltsyn Institute of Nuclear Physics, Lomonosov Moscow State University

Web Address: www.sinp.msu.ru

**Contact Information:**

Primary Technical Contact Information:

Name: Ivan Yashin

Organization and Position: Head of department

Address: Leninskie Gory, 1(2), Moscow, Russia, 119 991

Phone No.: +7(495)939-18-10

E-mail Address: ivn@eas.sinp.msu.ru

Alternate Technical Contact Information:

Name: Vasily Petrov

Organization and Position: \_\_\_\_\_

Address: Leninskie Gory, 1(2), Moscow, Russia, 119 991

Phone No.: +7(495)939-50-50

E-mail Address: vas.i.nas@gmail.com

Primary Science Contact Information:

Name: Mikhail Panasyuk

Organization and Position: Director of Institute

Address: Leninskie Gory, 1(2) Moscow, Russia, 119 991

Phone No.: +7(495)939-18-18

E-mail Address: panasyuk@sinp.msu.ru

Alternate Science Contact Information:

Name: Sergei Svertilov

Organization and Position: \_\_\_\_\_

Address: Leninskie Gory, 1(2), Moscow, Russia, 119 991

Phone No.: +7(495)939-51-60

E-mail Address: sis@coronas.ru

**Mission Specifics:**

Scientific or Engineering Objectives of Mission:

(specify below)

Purpose of the mission is studying of ultra high energy cosmic rays, gamma-ray bursts and transient luminous events in the upper atmosphere.

Role of Satellite Laser Ranging (SLR) for the Mission:

(specify below)

1. SLR will be used for additional control of disclosure rod-block of photodetectors by measuring the distance between two retroreflectors arrays;  
2. SLR will be used for the determination of spacecraft position.  
Two normal points sets should to be generated when two echoes are observed.

Anticipated Launch Date: April 2016 (preliminarily)

Expected Mission Duration: 3 years

Required Orbital Accuracy: \_\_\_\_\_

**Anticipated Orbital Parameters:**

Altitude (Min & Max for eccentric orbits): 478-500 km

Inclination: 97,272 degrees

Eccentricity: 0,00

Orbital Period: 94,2356 minutes

Frequency of Orbital Maneuvers: \_\_\_\_\_

**Mission Timeline:**

(example)

Should include when SLR is to start within the mission timeline, such as on insertion into orbit or "launch +N" days.

**Tracking Requirements:**

Tracking Schedule:  horizon-to-horizon  custom (specify: see comments Section II)

Spatial Coverage:  global ILRS network  custom (specify: \_\_\_\_\_)

Temporal Coverage:  full-time  custom (specify: see comments Section II)

Normal Point Bin Size (Time Span): 30 seconds

(Choose one from 5, 15, 30, 120 and 300 seconds. Justify if other bin size is required.)

(See the "Bin Size" of other satellites on the ILRS Web site at

[http://ilrs.gsfc.nasa.gov/missions/satellite\\_missions/current\\_missions/index.html](http://ilrs.gsfc.nasa.gov/missions/satellite_missions/current_missions/index.html) .)

**Orbit Generation:**

Prediction Center: \_\_\_\_\_

Prediction Technical Contact Information:

Name: \_\_\_\_\_

Organization and Position: \_\_\_\_\_

Address: \_\_\_\_\_

Phone No.: \_\_\_\_\_

E-mail Address: \_\_\_\_\_

Priority of SLR for POD: \_\_\_\_\_

Other Sources of POD:

GNSS  DORIS  Accelerometer  other (specify: \_\_\_\_\_)

## SECTION II: TRACKING RESTRICTIONS:

Several types of tracking restrictions have been required during some satellite missions. See [http://ilrs.gsfc.nasa.gov/satellite\\_missions/restricted.html](http://ilrs.gsfc.nasa.gov/satellite_missions/restricted.html) for a complete discussion.

- 1) Elevation restrictions: Certain satellites have a risk of possible damage when ranged near the zenith. Therefore a mission may want to set an elevation (in degrees) above which a station may not range to the satellite.
- 2) Go/No-go restrictions: There are situations when on-board detectors on certain satellites are vulnerable to damaged by intense laser irradiation. These situations could include safe hold position or maneuvers. A small ASCII file is kept on a computer controlled by the satellite's mission which includes various information and the literal "go" or "nogo" to indicate whether it is safe to range to the spacecraft. Stations access this file by ftp every 5-15 minutes (as specified by the mission) and do not range when the flag file is set to "nogo" or when the internet connection prevents reading the file.
- 3) Segment restrictions: Certain satellites can allow ranging only during certain parts of the pass as seen from the ground. These missions provide station-dependent files with lists of start and stop times for ranging during each pass.
- 4) Power limits: There are certain missions for which the laser transmit power must always be restricted to prevent detector damage. This requires setting laser power and beam divergence at the ranging station before and after each pass. While the above restrictions are controlled by software, this restriction is often controlled manually.

Many ILRS stations support some or all of these tracking restrictions. You may wish to work through the ILRS with the stations to test their compliance with your restrictions or to encourage additional stations that are critical to your mission to implement them.

The following information gives the ILRS a better idea of the mission's restrictions. Be aware that once predictions are provided to the stations, there is no guarantee that forgotten restrictions can be immediately enforced.

Can detector(s) or other equipment on the spacecraft be damaged or confused by excessive irradiation, particularly in any one of these wavelengths (532nm, 1064nm, 846nm, or 423nm)?

No     Yes (specify unsafe wavelengths: ) 432 nm

Are there times when the LRA (Laser Retroreflector Array) will not be accessible from the ground?

No     Yes (specify: \_\_\_\_\_)  
(If so, go/nogo or segmentation files might be used to avoid ranging an LRA that is not accessible.)

→ Skip the next questions and go directly to SECTION III if you answered "No" to both of the above questions.

Is there a need for an elevation tracking restriction?

No     Yes (What elevation (minimum to maximum in degrees)? max 80 degrees )

Is there a need for a go/no-go tracking restriction?

No     Yes (Explain the reason(s)? \_\_\_\_\_)

Is there a need for a pass segmentation restriction?

No     Yes (Explain the reason(s)? \_\_\_\_\_)

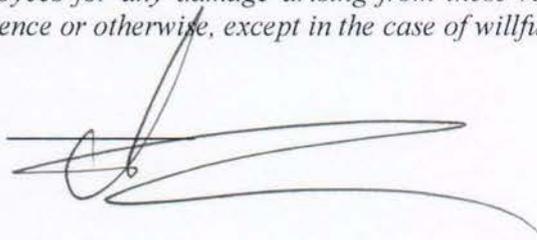
Is there a need for a laser power restriction?

No  
 Yes    (Under what circumstances? \_\_\_\_\_)  
(What is the maximum permitted power level **at** the satellite (nJ/cm<sup>2</sup>)? \_\_\_\_\_)  
(Is manual control of laser transmit power acceptable?     Yes     No)

For ILRS stations to range to satellites with restrictions, the mission sponsor must agree to the following statement:

*"The mission sponsor agrees not to make any claims against the station or station contractors or subcontractors, or their respective employees for any damage arising from these ranging activities, whether such damage is caused by negligence or otherwise, except in the case of willful misconduct."*

Please initial here to express agreement:



Other comments on tracking restrictions:  
(specify)

SC Lomonosov has a sensitive photodetectors for a wavelengths bellow 400 nm, which received the radiation count of 10 degrees from nadir direction. Tracking is NOT allowed when the SC is > 80 deg elevation regardless of the sunlit condition. Tracking is NOT allowed when the satellite is in the Earth's shadows regardless of the elevation angle. These restrictions must be implemented for the all wavelengths of SLR.

**SECTION III: RETROREFLECTOR ARRAY INFORMATION:**

A prerequisite for accurate reduction of laser range observations is a complete set of pre-launch parameters that define the characteristics and location of the LRA on the satellite. The set of parameters should include a general description of the array, including references to any ground-tests that may have been carried out, array manufacturer and whether the array type has been used in previous satellite missions. So the following information is requested:

Retroreflector Primary Contact Information:

Name: Andrey Sokolov

Organization and Position: "Research and Production Corporation "Precision Systems and Instruments"

Address: Russia Federation, Aviamotornya street, 53, 111 024

Phone No.: +7 (915) 152-43-50

E-mail Address: alsokolov@bk.ru

Array type:

- Single reflector  Spherical  Hemispherical/Pyramid  Planar  
 other (specify: \_\_\_\_\_ )

Attach a diagram or photograph of the satellite that shows the position of the LRA, at the end of this document.

Attached

Attach a diagram or photograph of the whole LRA at the end of this document.

- Attached  Same as above, Not attached (acceptable only for a cannonball satellite)

Array manufacturer:

"Research and Production Corporation "Precision Systems and Instruments"

Link (URL and/or reference) to any ground-tests that were carried out on the array:

\_\_\_\_\_  
\_\_\_\_\_

Has the LRA design and/or type of cubes been used previously?

- No  Yes (List the mission(s): \_\_\_\_\_ )

For accurate orbital analysis it is essential that full information is available in order that the 3-dimensional position of the satellite center of mass may be referred to the location in space at which the laser range measurements are made. To achieve this, the 3-D location of the LRA phase center must be specified in a satellite-body-fixed reference frame with respect to the satellite's mass center. In practice this means that the following parameters must be available at 1 mm accuracy or better.

Define the satellite-body-fixed XYZ coordinates (i.e. origin and axes) on the spacecraft:  
(specify) (add a diagram)

See attachment

Relate the satellite-body-fixed XYZ coordinates to a Celestial/Terrestrial/Solar Reference Frame including the attitude control policy:

(specify) (add a diagram if applicable)

The 3-D location of the satellite's mass center in satellite-body-fixed XYZ coordinates is:

- Always fixed at (0, 0, 0)
- Always fixed at ( \_\_\_\_\_ , \_\_\_\_\_ , \_\_\_\_\_ ) in mm
- Time-varying by approximately ( \_\_\_\_\_ ) mm during the mission lifetime.  
Will a time-variable table of the mass center location be available on the web?
  - No
  - Yes (URL: \_\_\_\_\_ )

The 3-D location (or time-variable range) of the phase center of the LRA in the satellite-body-fixed XYZ coordinates:

( \_\_\_\_\_ , \_\_\_\_\_ , \_\_\_\_\_ ) in mm

The following information on the corner cubes must also be supplied.

The XYZ coordinates referred to in the following are given in:

- Satellite-body-fixed system (same as above)
- LRA-fixed system (specify below)  
(specify the origin and orientation) (add a diagram if applicable)

List the position (XYZ) of the center of the front face of each corner cube, and the orientation (two angles or normal vector) and the clocking (horizontal rotation) angle of each corner cube. Note that the angles should be clearly defined.

- Attached at the end of this document
- Listed here (acceptable for small number (20 or less) of corner cubes)  
(specify) (add a diagram if applicable)

Is the corner cube recessed in its container (i.e. can the container obscure a part of the corner cube)?

- No
- Yes (specify below)  
(specify) (add a diagram)

The size of each corner cube: Diameter (15) mm Height (11,25) mm

The material from which the cubes are manufactured (e.g. quartz):

BK-7

The refractive index of the cube material

= 1.51947 for wavelength  $\lambda = 0.532$  micron

= see attachment as a function of wavelength  $\lambda$  (micron):

The group refractive index of the cube material, as a function of wavelength  $\lambda$  (micron):

= 1.54894 for wavelength  $\lambda = 0.532$  micron

= see attachment as a function of wavelength  $\lambda$  (micron):

Dihedral angle offset(s) and manufacturing tolerance (in arcseconds):

See attachment

Radius of curvature of front surfaces of cubes:

Not applied     Yes (specify: \_\_\_\_\_)

Flatness of cubes' surfaces:

1/8λ

Back-face coating:

Uncoated     Coated (specify the material: Reflecting surface of cube corner coated aluminium)

Other Comments:

(specify) (add a diagram if applicable)

**SECTION IV: MISSION CONCURRENCE**

As an authorized representative of the "Lomonosov" mission, I hereby request and authorize the ILRS to track the satellite described in this document.

Name (print): Mikhail Panasyuk

Organization and Position: SINP MSU, director

Signature: \_\_\_\_\_

Date: March 2, 2016  


Send form to: ILRS Central Bureau  
c/o Carey Noll  
NASA GSFC  
Code 690  
Greenbelt, MD 20771  
USA  
301-614-6542 (Voice)  
301-614-6015 (Fax)  
Carey.Noll@nasa.gov

**SECTION V: ATTACHMENT(S)**